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PLOTIT-METHOD OF INTERACTIVELY PLOTTING  
INPUT DATA FOR THE VORLAX COMPUTER PROGRAM

(NASA C -158896) PLOTIT-METHOD OF  
INTERACTIVELY PLOTTING INPUT DATA FOR THE  
VORLAX COMPUTER PROGRAM (Vought Corp.,  
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## SUMMARY

A method of plotting the geometric input to the VORLAX computer program by means of an interactive remote computer terminal is described. The software consists of a procedure file and two programs and was developed for use with the Langley Research Center computer system. The programs and procedure file are described and a sample execution is presented.

## INTRODUCTION

The VORLAX computer program uses a sparse set of geometric input data to describe the aircraft configuration being analyzed. The geometry of the configuration can at times become very complex, and it is necessary to plot the configuration resulting from the data in order to ascertain its accuracy. The procedure file and two computer programs described herein provide a method for plotting this data at an interactive graphics terminal. A sample execution of the procedure is presented.

Use of this method allows the configuration to be plotted with any combination of roll, pitch, or yaw angles. Three independent forms of data display are available, and these may be specified in any combination. These are: (1) configurations with or without camber, (2) configurations showing only major panels or only minor panels, and (3) configurations with or without control points plotted. Any section of the plot may be enlarged for examination in greater detail.

The procedure file and computer programs to plot the VORLAX input data have been written to be used in the Langley Research Center computer system which provides a Network Operating System (NOS) and a Tektronix Plot 10 package. Langley Research Center users will find the procedure file and computer programs in mass storage as public files in the catalog of user number 214737C.

## DISCUSSION

### DESCRIPTION OF PROCEDURE FILES AND PROGRAMS

A method has been developed for plotting the VORLAX input data which consists of a procedure file and two computer programs. The procedure file PLOTIT and the computer programs READS and PLOTS are described in the following. A sample execution is provided which includes illustrations of the displays.

#### Procedure File PLOTIT

Procedure file PLOTIT (Appendix A) is used to simplify the plotting procedure. PLOTIT first gets the desired data file and renames it TAPE1. It then gets the binary form of program READS, which reads the input data and prepares files suitable for the subsequent operations. PLOTIT then gets and executes the binary form of PLOTS, which performs the aircraft geometry plotting.

#### Program READS

Program READS (Appendix B) reads the VORLAX input data from a disc file named TAPE1 and determines the necessary scaling factor in order for the aircraft drawing to fit on the screen.

The data for each of the panels are then read, and the coordinates of a set of points that describe the panels are calculated. These coordinates are stored on three disc files named TAPE3, TAPE4, and TAPE5. These data files will be read by program PLOTS.

Program READS provides the user with two methods of representing the aircraft. The first method displays the aircraft with camber as shown in figure 1(a). The second method shows the aircraft without camber. Figure 1(b) is the aircraft of figure 1(a) without camber.

#### Program PLOTS

Program PLOTS (Appendix C) reads the data produced by program READS. This program offers the user several variations in the display which are described

in the following sections.

Rotation of the Aircraft. - The initial position of the aircraft is a side view with the nose to the left. From this position, it is possible to rotate the aircraft first in roll, then in pitch, and finally in yaw. Positive angles are defined as follows: roll, right wing down; pitch, nose up; and yaw, nose right.

Additional Independent Variations. - PLOTS offers two additional independent variations in the display. These variations are: (1) with or without subpaneling and (2) with or without control points. Examples of these variations are shown in figures 1(c) through 1(f). Figure 1(c) shows the aircraft without subpaneling or control points; figure 1(d) has subpaneling added; in figure 1(e) control points only have been added; and in figure 1(f) control points and subpaneling have been added. All of the plots are constructed using orthographic projection.

Recovery from Input Errors. - If an error has been made in specifying the input parameters, the execution of PLOTS can be stopped. There are three ways to stop PLOTS while it is executing. These are: (1) enter a value greater than 360 for roll angle, (2) answer any of the questions with STOP, or (3) stop the program while it is plotting by interrupting it with the break key and then entering an S.

Program PLOTS can be restarted at the beginning by sending the command PLOTB to the computer. PLOTB will also restart the program if it is stopped for any other reason.

#### SAMPLE EXECUTION

Figure 2 shows a sample execution of PLOTIT at a remote terminal. The first command gets the procedure file PLOTIT.

```
GET,PLOTIT/UN=214737C
```

The second command initiates the execution of the procedure file.

CALL,PLOTIT(T=VORLAXX)

VORLAXX is the name of the file on which the VORLAX data deck has been stored for this example. The file name in the calling statement can be any name which corresponds to a file on which VORLAX data is stored.

The first question asked by the computer deals with camber in the panels.

DO YOU WANT CAMBER IN THE PANELS ?  
TRUE OR FALSE

If camber is desired in the panels, type in TRUE, otherwise type in FALSE and the camber will be set equal to zero.

The next three questions asked by the computer are concerned with the desired roll, pitch, and yaw angles of the configuration, and are as follows.

INPUT THE ROLL ANGLE FOR THE AIRCRAFT  
(DEG), > 360 TO STOP.  
PITCH ANGLE  
YAW ANGLE

The angles desired in degrees, are typed in after the questions. If termination of the program is desired, a value greater than 360 may be typed in for the roll angle.

The next two questions are concerned with the desirability of displaying paneling and control points. The two questions are:

SUBPANELING ? TRUE OR FALSE  
CONTROL POINTS ? TRUE OR FALSE

If these questions are answered TRUE, the subpaneling and control points are incorporated into the plots. If they are answered FALSE, then these quantities are deleted. The plot resulting from the input in figure 2 is presented in figure 3.

If a certain section of the plot needs to be enlarged in order to examine the plotted data more closely, this may be accomplished at the terminal. When the computer has finished plotting, it will print the following statement:

FOR ENLARGEMENT INPUT YES

At this point a hard copy can be made if desired. If any reply except YES is typed in, the computer will ask for a new set of angles. If YES is typed in, the graphics cursor (cross hairs) will appear. The cursor should then be located at the lower left corner [fig. 4(a)] of a rectangular region to be enlarged. A non-control keyboard character should be pressed. This will cause the cursor to disappear. The carriage return is then pressed. This sequence sends the coordinates of the first corner to the computer. The graphics cursor will reappear and should be relocated to the upper right corner of the desired rectangular region [fig. 4(b)]. A non-control keyboard character and the carriage return are then pressed as for the first corner. An enlargement of the region defined by these positions of the graphics cursor is shown in figure 4(c).

When the plot is finished, the computer will again print

·FOR ENLARGEMENT INPUT YES

This allows a further enlargement of a section of the plot if desired.

#### CONCLUDING REMARKS

A plotting routine, PLOTIT, has been developed for plotting the input data for the VORLAX computer program. This program allows the user to plot geometric input data interactively at a remote graphics terminal and thereby ascertain very rapidly whether or not the data is correct.

The routine consists of two programs and a procedure file. These have been designed for use on the Control Data Corporation computer system with a Network Operating System (NOS) and a Tektronix Plot 10 graphics package at the NASA Langley Research Center.

#### REFERENCES

1. Miranda, Luis R.; Elliott, Robert D.; and Baker, William M.: A Generalized Vortex Lattice Method for Subsonic and Supersonic Flow Applications. NASA CR-2865, December 1977.

## APPENDIX A

### PROCEDURE FILE PLOTIT

This procedure file gets the binary form of READS(READB) and executes it, then gets the binary form of PLOTS (PLOTB) and executes it.

```
PLOTIT.  
RETURN,TAPE3,TAPE4,TAPE5.  
GET,TAPE1=T.  
GET,READB/UN=214737C.  
READB.  
RETURN,READB.  
GET,PLOTB/UN=214737C.  
PLOTB.  
EXIT.
```



## APPENDIX B

## SOURCE LISTING OF PROGRAM READS

```

C
C
PROGRAM READS(INPUT,OUTPUT,TAPE1,TAPE4,TAPE2=OUTPUT,TAPE3,TAPE5)
  THIS PROGRAM READS THE INPUT DATA FOR THE VORLAX PROGRAM
  SO IT CAN BE PLOTTED.
  DIMENSION TITLE(8)
  COMMON/BLOCK/XOFFSET
  LOGICAL CAMBER
  REAL LENGTH
  READ(1,100)TITLE
  READ(1,103)LAX,LAY
  READ(1,101)
  READ(1,102)NPAN,WSPAND
  CALL SIZES(NPAN,LENGTH,XOFFSET,WSPAN)
  IF(WSPAND.GT.WSPAN)WSPAN=WSPAND
  TESTRTD=LENGTH/WSPAN
  IF(LENGTH.GT.WSPAN)WSPAN=LENGTH
  REALY IS THE WIDTH OF THE PLOTTING SURFACE.
  WRITE(3)NPAN,TITLE,TESTRTD
  REALY=10.0
  REALY=REALY-.20
  SCALE=REALY/WSPAN
  XOFFSET=-(XOFFSET+WSPAN*.5)
  WRITE(2,104)
  READ 105,CAMBER
  DO 200 I=1,NPAN
    200 CALL PANLRED(SCALE,LAX,LAY,CAMBER)
  100 FORMAT(8A10)
  101 FORMAT(//)
  102 FORMAT(I2,48X,F10.0)
  103 FORMAT(11X,I1,9X,I1)
  104 FORMAT("DO YOU WANT CAMBER IN THE PANELS ? ",/, "TRUE OR FALSE")
  105 FORMAT(L7)
  STOP
  END

```

# APPENDIX B. - Continued

34	READS	
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```

SUBROUTINE PANLRED(SCALE,LAX,LAY,CAMBER)
COMMON/BLOCK/XOFFSET
COMMON/BLOCK/NAP,XAF(50),ZC(2,50),CORD1,CORD2,C01SIN,C02SIN
COMMON/CANDS/SIN1,SIN2,C0S1,C0S2,DELTA
COMMON/ROJ1/ROJ,CROSSIZ
LOGICAL CAMBER
INTEGER PRD,RNCV

      THIS SUBROUTINE READS THE VORLAX DATA FOR A PANEL,EACH TIME IT
      IS CALLED.  THE COORDINATES OF A SET OF POINTS THAT DESCRIBE
      THE PANEL ARE CALCULATED AND STORED ON DISC.

      DIMENSION X(2),Y(2),Z(2),CORD(2),RLE(2)
      COMMON/PHIRO/ PHI(100),RO(100),SINE(100),COSINE(100)
      COMMON VORS(3,500)
      COMMON/PI/PIE
      COMMON/TWIST/AINC1,DAINC
      REAL K
      DATA PIE/3.14159/,CROSSIZ/.02/

      THIS SECTION READS THE VORLAX DATA CARDS FOR A PANEL.

      DO 200 I=1,2
200  READ(1,100)X(I),Y(I),Z(I),CORD(I)
      DO 250 I=1,2
250  X(I)=(X(I)+XOFFSET)*SCALE
      Y(I)=Y(I)*SCALE
      Z(I)=Z(I)*SCALE
      CORD(I)=CORD(I)*SCALE
      READ(1,101)TVOR,TNCV,POL
      RNCV=INT(TNCV)
      NVOR=INT(TVOR)
      NVOR1=NVOR+1
      IF(POL.LE.360.)GO TO 1
      READ(1,102)(PHI(N),RO(N),N=1,NVOR1)
      DO 251 N=1,NVOR1
251  PHI(N)=PHI(N)*3.14159/180.
      RO(N)=RO(N)*SCALE
  
```

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## APPENDIX B. - Continued

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VORS(2,I2)=YI+DLTAY*RATIO
VORS(3,I2)=Z1+DELTAZ*RATIO
XCOR0=CDICOS+(DELTA CO)*RATIO
XX=XI+RATIO*(DELTAX)
VORS(3,I2)=VORS(3,I2)+ZZO+ZZ*RATIO
VORS(1,I2)=XX+PCORD*XCORD+XOO+RATIO*XERP
501 IF(I2.EQ.500)CALL SAVE(IFUG,3,IDUM,2)
CALL INTERP(L,O,ZZO1,ZZ1,XOO1,XERPL)
DO 300 I=1,NVORI
I2=NVRN+NVIOR1+I-1+IFUG
IF(LAY.EQ.O)RATIO=(I-1)/NVOR
IF(LAY.EQ.O)KATI0=.5*(1.-COS(PIE*(I-1.)/(NVOR)))
VORS(2,I2)=YI+DELTAY*RATIO
VORS(3,I2)=ZZO1+RATIO*(ZZI+DELT AZ)+ZI
VORS(1,I2)=(DELT AX+DELTACU+XERPL)*RATIO+XI+CDICOS+XOO1
300 IF(I2.EQ.500)CALL SAVE(IFUG,3>IDUM,2)
CALL SAVE(IFUG,3,NVRNPRD,3)
C * * * * *
C THIS SECTION OF THE PROGRAM CALCULATES THE LOCATION OF THE
C CONTROL POINTS FOR A FLAT PANEL SUCH AS A WING. IT THEN
C PLACES AN "X" ON EACH POINT.
PRD=NVOR*NRCV
NNO=(RNCV-1)*NVOR+1
NNN=PRD#4
ISETS=ISETSF(NNN)
WRITE(4) NVOR,RNCV,PRD,NNN,NNN,NNO,ISETS
CALL SAVE(IFUG,4>IDUM,1)
DO 1001 I=1,NNO,NVOR
K=(I-1)/NVOR+1
IF(LAX.EQ.O)PCORD=.5*(1.-COS(K*PIE/RNCV))
IF(LAX.EQ.O)PCORD=(4.*K-1.)/(N4)
CALL INTERP(PCORD,ZZO,ZZ,XOO,XERP)
(RAT=SQRT(DELT AY*DELTAY+(DELT AZ+ZZ)**2.)
V1=CROSSIZ*DELTAY/YRAT
V2=CROSSIZ*(DELT AZ+ZZ)/YRAT
DO 1001 J=1,NVOR

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# APPENDIX B. - Continued

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J2=(I+J-2)*4+IFUG
J21=J2+1
J22=J2+2
J23=J2+3
J24=J2+4
IF(LAY.EQ.1)RATIO=(J-.5)/NVOR
IF(LAY.EQ.0)RATIO=.25*(2.-COS(PIE*J/NVOR)-COS(PIE*(J-1.)/NVOR))
CALL ANGLE(RATIO,WS,WC)
XCORD=CO1COS+DELTACO*RATIO
TEMX=X1+RATIO*(DELTAX+XERP)+PCORD*XCORD+X00
VORS(1,J21)=TEMX-CROSSIZ*WC
VORS(1,J22)=TEMX+CROSSIZ*WC
VORS(1,J23)=VORS(1,J24)=TEMX
TEMY=Y1+DELTAY*RATIO
VORS(2,J21)=VORS(2,J22)=TEMY
TEMZ=Z1+DELTAZ*RATIO
TEMZ=TEMZ+ZZG+ZZ*RATIO
VORS(3,J21)=TEMZ-CROSSIZ*WS
VORS(3,J22)=TEMZ+CROSSIZ*WS
VORS(3,J23)=TEMZ-V2
VORS(3,J24)=TEMZ+V2
VORS(2,J23)=TEMY-V1
VORS(2,J24)=TEMY+V1
IF(J24.EQ.500)CALL SAVE(IFUG,4,IDUM,2)
1001 CONTINUE
CALL SAVE(IFUG,4,NNN,3)
1010 CONTINUE
RETURN
C * * * * * THIS SECTION OF THE PROGRAM IS FOR CURVED MAJOR PANELS.
C * * * * *
C
5 YNOT=Y(1)-RO(1)*COS(PHI(1))
ZNOT=Z(1)-RO(1)*SIN(PHI(1))
TEMP2=X1+CORD1
C THIS CALCULATES THE SCALING FACTORS FOR THE RADIUS VECTORS
C FROM AREA RATIOS.
IF(NAP.LE.2)GO TO 90
DO 89 J=1,NAP

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## APPENDIX B. - Continued

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      89 ZC(2,J)=SQRT(ZC(2,J)*.01)
      C
      C      THIS SECTION CALCULATES THE LOCATION OF A SET OF POINTS THAT
      C      DESCRIBE A CURVED MAJOR PANEL SUCH AS A FUSELAGE OR A NACELLE
      C      WITH SUBPANELING.
      C
      90 CALL INTERP2(0.0,XSHFT1,SKAL1,DM,DM,ZSHFT1,DM,9H NO POINT)
      CALL SAVE(IFUG,5,IDUM,1)
      ISETS=ISETSF(NVRNPRD)
      WRITE(5)NVRN,RNCV,NVRNPRD,ISETS
      DO 1333 I=1,NVOR1
      R=RQ(I)
      FI=PHI(I)
      VORS(1,I)=X1
      VORS(2,I)=R*COS(FI)*SKAL1+YNQT
      1333 VORS(3,I)=R*SIN(FI)*SKAL1+ZNQT+ZSHFT1
      C
      DO 1501 I=NVOR1,NVRN,NVOR1
      IF(LAX.EQ.1)PCORD=(I/NVOR1+.3.)/N4
      IF(LAX.EQ.0)PCORD=.5*(1.-COS((2.*I/NVOR1-1.)*PIE/N2))
      CALL INTERP2(PCORD,XSHFT,SKAL,DM,DM,ZSHFT,DM,9H NO POINT)
      XTEMP=X1+XSHFT
      DO 1501 J=1,NVOR1
      I2=I+J+IFUG
      R=RQ(J)*SKAL
      FI=PHI(J)
      VORS(1,I2)=XTEMP
      VORS(2,I2)=R*COS(FI)+YNQT
      VORS(3,I2)=R*SIN(FI)+ZNQT+ZSHFT
      1501 IF(I2.EQ.500)CALL SAVE(IFUG,5,IDUM,2)
      CALL INTERP2(1.0,XSHFT2,SKAL2,DM,DM,ZSHFT2,DM,9H NO POINT)
      DO 1300 I=1,NVOR1
      I2=NVRN+NVOR1+I-1+IFUG
      R=RQ(I)
      FI=PHI(I)
      VORS(1,I2)=T*MP2
      VORS(2,I2)=R*SKAL2*COS(FI)+YNQT
      VORS(3,I2)=R*SKAL2*SIN(FI)+ZNQT+ZSHFT2

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# APPENDIX B. - Continued

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1300 IF(I2.EQ.500)CALL SAVE(IFUG,5,IDUM,2)
      CALL SAVE(IFUG,5,NVRNPRD,3)
C
C      THIS SECTION CALCULATES THE LOCATION OF A SET OF POINTS THAT
C      DESCRIBE A CURVED MAJOR PANEL, WITHOUT SUBPANELING.
      NVRN=(NAP)*(NVOR+1)
      NVRNPRD=(NAP)*(NVOR+1)
      ISETS=ISETSF(NVRNPRD)
      WRITE(3)NVOR,NAP,PDL,IQUANT,NVRNPRD,ISETS
      CALL SAVE(IFUG,3,NVRNPRD,1)
      IF(NAP.LE.2)GO TO 556
      DO 555 I=1,NVRN,NVOR1
        II=(I)/NVOR1+1
        XTEMP=X1+CORD1*XAF(II)
        SKAL=ZC(2,II)
        DO 555 J=1,NVOR1
          R=RO(J)*SKAL
          FI=PHI(J)
          I2=I+J-1+IFUG
          VORS(1,I2)=XTEMP
          VORS(2,I2)=P*COS(FI)+YNOT
          VORS(3,I2)=P*SIN(FI)+ZNOT+CORD1*.01*ZC(1,II)
555   IF(I2.EQ.500)CALL SAVE(IFUG,3,IDUM,2)
          IF(NAP.GT.2)GO TO 5555
556   NAP=2
          NVRNPRD=NAP*NVOR1
          DO 5554 J=1,NVOR1
            J2=J+NVOR1
            VOPS(1,J)=X1
            VOPS(1,J2)=X1+CORD1
            VOPS(2,J)=VORS(2,J2)-YNOT+RO(J)*COS(PHI(J))
            VOPS(3,J)=VORS(3,J2)-ZNOT+RO(J)*SIN(PHI(J))
5554   VOPS(3,J)=VORS(3,J2)-ZNOT+RO(J)*SIN(PHI(J))
5555   CALL SAVE(IFUG,3,NVRNPRD,3)
C
C      THIS SECTION CALCULATES THE LOCATION OF THE CONTROL POINTS FOR
C      A CURVED MAJOR PANEL. IT THEN PLACES AN "X" ON EACH POINT.
      POJ=PO(1)
      DO 2010 N=1,NVOR

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# APPENDIX B. - Continued

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300 READS
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      AZO=RO(N)*SIN(PHI(N))
      AZ1=RO(N+1)*SIN(PHI(N+1))
      AYO=RO(N)*COS(PHI(N))
      AY1=RO(N+1)*COS(PHI(N+1))
      DAZ=AZO-AZ1
      DAY=AYO-AY1
      H=SQRT(DAZ*DAZ+DAY*DAY)
      SINE(N)=DAZ/H
      COSINE(N)=DAY/H
      R1=RO(N)
      R2=RO(N+1)
      F1=PHI(N)
      F2=PHI(N+1)
      RO(N)=(R1*SIN(F1)+R2*SIN(F2))*0.5
      PHI(N)=(R1*COS(F1)+R2*COS(F2))*0.5
      PRD=NVDOR*RNCV
      NNO=(RNCV-1)*NVOR+1
      NNN=PRD*4
      ISETS=ISETSF(NNN)
      WRITE(4)NVOR,RNCV,PRD,NNN,NNN,NNO,ISETS
      CALL SAVE(IFUG,4,IDUM,1)
      DO 2001 I=1,NNO,NVDOR
      K=(I-1)/NVOR+1
      IF(LAX.EQ.0)PCORD=.5*(1.-COS(K*PIE/RNCV))
      IF(LAX.EQ.1)PCORD=(4.*K-1.)/N4
      CALL INTERP2(PCORD,TEMX,SKAL,DSKL,CROSSX,CAMB,DZ,6HPOINTS)
      TEMX=TEMX+X1
      DO 2001 J=1,NVDOR
      J2=(I+J-2)*4+IFUG
      J21=J2+1
      J22=J2+2
      J23=J2+3
      J24=J2+4
      VS=SINE(J)*CRDSSIZ
      HS=COSINE(J)*CROSSIZ
      TEMY=PHI(J)*SKAL+YNOT
      TY2=DSKL*PHI(J)
      TEMZ=RO(J)*SKAL+ZNOT+CAMB

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# APPENDIX B. - Continued

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T22=RO(J)*DSKL+DZ
VORS(1,J21)=TEMX-CROSSX
VORS(1,J22)=TEMX+CROSSX
VORS(1,J23)=TEMX
VORS(1,J24)=TEMX
VORS(2,J21)=TEMX-TY2
VORS(2,J22)=TEMX+TY2
VORS(2,J23)=TEMX-HS
VORS(2,J24)=TEMX+HS
VORS(3,J21)=TEMX-TZ2
VORS(3,J22)=TEMX+TZ2
VORS(3,J23)=TEMX-VS
VORS(3,J24)=TEMX+VS
IF(J24.EQ.500)CALL SAVE(IFUG,4,IOUM,2)
2001 CONTINUE
CALL SAVE(IFUG,4,NNN,3)
RETURN
END

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338 READS  
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SUBROUTINE INTERP(P,ZA,Z,XA,X)
  THIS SUBROUTINE DOES THE INTERPOLATION FOR CAMBER OF A FLAT
  MAJOR PANEL.
  COMMON/BLOKO/NAP,XAF(50),ZC(2,50),CORD1,CORD2,COS1N,COS2SIN
  COMMON/CANDS/SIN1,SIN2,COS1,COS2,DELTA
  COMMON/TWIST/AINC1,DAINC
  COMMON/SLOPE/ANGL,DANG
  IF(NAP.LE.2.OR.DELTA.EQ.0.0)GO TO 3
  DO 1 I=2,NAP
    1 IF(P.LE.XAF(I))GO TO 2
    2 DAF=XAF(I)-XAF(I-1)
    DZC1=ZC(1,I)-ZC(1,I-1)
    DZC2=ZC(2,I)-ZC(2,I-1)
    A1=ATAN(DZC1/DAF*.01)
    A2=ATAN(DZC2/DAF*.01)
    DAA=A2-A1
    PC=(P-XAF(I-1))/DAF
    CAM91=(DZC1*PC+ZC(1,I-1))*CORD1*.01

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# APPENDIX B. - Continued

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CAMB2=(DZC2\*PC+ZC(2,I-1))\*CORD2\*.01  
GO TO 4  
3 CAMB1=CAMB2=A1=DAA=0.0  
4 DANG=DAA+DAINC  
ANGL=AINC1+A1  
ZA=CAMB1\*COS1+P\*CD1SIN  
ZB=CAMB2\*COS2+P\*CD2SIN  
Z=ZB-ZA  
XA=-CAMB1\*SIN1  
XB=-CAMB2\*SIN2  
X=XB-XA  
RETURN  
END

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SUBROUTINE INTERP2(P,X,S,DS,CROS,Z,DZ,TEST)  
THIS SUBROUTINE DOES THE INTERPOLATION FOR THE RADIUS VECTORS  
AND THE CAMBER OF A CURVED MAJOR PANEL.  
COMMON/BLK0/NAP,XAF(50),ZC(2,50),CORD1,CORD2,CD1SIN,CD2SIN  
COMMON/ROJ1/ROJ,CROSSIZ  
X=P\*CORD1  
IF(NAP.LE.2)GO TO 3  
DO 1 I=2,NAP  
1 IF(P.LE.XAF(I))GO TO 2  
2 DAF=XAF(I)-XAF(I-1)  
DZC1=ZC(1,I)-ZC(1,I-1)  
DZC2=(ZC(2,I)-ZC(2,I-1))  
PC=(P-XAF(I-1))/DAF  
S=ZC(2,I-1)+DZC2\*PC  
Z=ZC(1,I-1)+DZC1\*PC  
Z=Z\*.01\*CORD1  
IF(TEST.NE.6HPOINTS)RETURN  
DRAD=ROJ\*DZC2  
DELX=DAF\*CORD1  
COSINE=DELX/SQRT(DRAD\*DRAD+DELX\*DELX)  
CROS=COSINE\*CROSSIZ  
P2=CROS/CORD1

C  
C

# APPENDIX B. - Continued

READS 410  
 READS 411  
 READS 412  
 READS 413  
 READS 414  
 READS 415  
 READS 416  
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 READS 418  
 READS 419  
 READS 420  
 READS 421

PC2=(P+P2-XAF(I-1))/DAF  
 S2=ZC(2,I-1)+DZC2\*PC2  
 Z2=ZC(1,I-1)+DZC1\*PC2  
 Z2=Z2+CORD1\*.01  
 DZ=Z2-Z  
 DS=S2-S  
 RETURN  
 3 S=S2+1.0  
 Z=DS=DZ=0.0  
 CROS=CROSSIZ  
 RETURN  
 END

READS 422  
 READS 423  
 READS 424  
 READS 425  
 READS 426  
 READS 427  
 READS 428  
 READS 429  
 READS 430  
 READS 431

SUBROUTINE ANGLE(R,S,C)  
 THIS SUBROUTINE IS USED FOR PLACING X'S ON CONTROL POINTS,  
 ON FLAT MAJOR PANELS.  
 COMMON/SLOPE/ANGL,DANG  
 ANG=ANGL+R\*DANG  
 S=SIN(ANG)  
 C=COS(ANG)  
 RETURN  
 END

READS 432  
 READS 433  
 READS 434  
 READS 435  
 READS 436  
 READS 437  
 READS 438  
 READS 439  
 READS 440  
 READS 441

SUBROUTINE ZEROZC1(POL)  
 THIS SUBROUTINE REMOVES CAMBER FROM THE MAJOR PANELS.  
 COMMON/BLOK0/NAP,XAF(50),ZC(2,50),CORD1,CORD2,CD1SIN,CD2SIN  
 DO 1 I=1,2  
 IF(I.EQ.2,A.POL.GE.360.)RETURN  
 DO 1 J=1,NAP  
 1 ZC(I,J)=0.0  
 RETURN  
 END

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# APPENDIX B. - Continued

C	FUNCTION ISETSF(N)	442
C	THIS FUNCTION DETERMINES THE NUMBER OF DATA SETS NEEDED TO	443
	TC DESCRIBE A MAJOR PANNEL.	444
	R=1.0/500.*N	445
	ISETSF=INT(R)	446
	IF(ISETSF.LT.R)ISETSF=ISETSF+1	447
	RETURN	448
C	END	449
		450
C	SUBROUTINE SAVE(FUG,FILE,PROD,TEST)	451
C	THIS SUBROUTINE MAKES IT POSSIBLE TO REDUCE CORE REQUIREMENTS	452
C	BY REDUCING ARRAY SIZE.	453
	INTEGER FUG,FILE,PNTS,TEST,PROD	454
	COMMON VORS(3,500)	455
	IF(TEST.EQ.2)GO TO 2	456
	IF(TEST.EQ.3)GO TO 3	457
	FUG=0	458
	RETURN	459
	2 FUG=FUG-500	460
	WRITE(FILE)((VORS(I,J),I=1,3),J=1,500)	461
	RETURN	462
	3 PNTS=MOD(PROD,500)	463
	IF(PNTS.EQ.0)RETURN	464
	WRITE(FILE)((VORS(I,J),I=1,3),J=1,PNTS)	465
	RETURN	466
C	END	467
		468
C **	SUBROUTINE SIZES(NPAN,LENGTH,XOFFSET,WIDTH)	469
C **	DETERMINES THE LENGTH, WIDTH, STARTING POINT OF THE	470
	AIRCRAFT.	471
	REAL LENGTH	472
	NPAN2=NPAN-1	473
	CALL FIND(LENGTH,XOFFSET,WIDTH)	474
	IF(NPAN.EQ.1)GO TO 8	475

# APPENDIX B. - Continued

476 READS  
477 READS  
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480 READS  
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488 READS

```

DO 7 I=1,NPAN2
  CALL FIND(BIG,SMALL,WIDE)
  IF(SMALL.LT.XOFFSET)XOFFSET=SMALL
  IF(WIDE.GT.WIDTH)WIDTH=WIDE
7  IF(BIG.GT.LENGTH)LENGTH=BIG
  WIDTH=WIDTH*2.0
  LENGTH=LENGTH-XOFFSET
8  REWIND 1
  READ(1,5)
5  FORMAT(////)
  RETURN
  END

```

C

489 READS  
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```

SUBROUTINE FIND(BIG,SMALL,WIDE)
  DETERMINES THE STARTING POINT, END POINT, AND
  DISTANCE FROM THE AXIS FOR A PANEL.
  COMMON/BLK0/NAP,XAF(50),ZC(2,50),CORD1,CORD2,CD1SIN,CD2SIN
  COMMON/PHIR0/ PHI(100),RO(100),SINE(100),COSINE(100)
  READ(1,100)X1,Y1,Z1,CORD1
  READ(1,100)X2,Y2,Z2,CORD2
  Y1=ABS(Y1)
  Y2=ABS(Y2)
  READ(1,101)TVOR,TNCV,PDL
  NVOR1=INT(TVOR)+1
  IF(PDL.LE.360.)GO TO 1
  READ(1,102)(PHI(N),RO(N),N=1,NVOR1)
1  READ(1,103)AINC1,AINC2,ITS,NAP,IQUANT,ISYNT,NPP
  IF(ISYNT.NE.0)READ(1,104)
  IF(NAP.LE.2)GO TO 2
  READ(1,102)(XAF(I),I=1,NAP)
  IF(ITS.EQ.0.OR.PDL.GE.360.)GO TO 3
  READ(1,105)R
3  READ(1,102)(ZC(1,I),I=1,NAP)
  IF(ITS.EQ.0.OR.PDL.GT.360.)GO TO 4
  READ(1,105)R

```

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# APPENDIX B. - Concluded

READS	511
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READS	529

```

4 READ(1,102)(ZC(2,I),I=1,NAP)
2 CONTINUE
100 FORMAT(4F10.0)
101 FORMAT(2F10.0,10X,F10.0)
102 FORMAT(8F10.0)
103 FORMAT(2F10.0,12,8X,12,9X,11,9X,11,9X,11)
104 FORMAT(/)
105 FORMAT(F10.0)

      SMALL=X2
      IF(X1.LT.SMALL)SMALL=X1
11  X1=X1+CORD1
      X2=X2+CORD2
      BIG=X2
      IF(X1.GT.BIG)BIG=X1
      WIDE=Y1
      IF(Y2.GT.WIDE)WIDE=Y2
      RETURN
      END

```

C

# APPENDIX C

## SOURCE LISTING OF PROGRAM PLOTS

C	PROGRAM PLOTS(INPUT,OUTPUT,TAPE2=OUTPUT,TAPE3,TAPE4,TAPE5)	PLOTS	1
	THIS INITIALIZES THE PLOTTING ROUTINES AND CALLS PLOTSPAN.	PLOTS	2
	CALL INITT(120)	PLOTS	3
	CALL TERM(3,4096)	PLOTS	4
	CALL CHRISZ(4)	PLOTS	5
	CALL PLOTSPAN	PLOTS	6
	CALL FINITT(0,0)	PLOTS	7
	STOP	PLOTS	8
	END	PLOTS	9
C		PLOTS	10
	SUBROUTINE PLOTSPAN	PLOTS	11
	COMMON VORS(3,500)	PLOTS	12
	COMMON /PRAMS/NVOR,NVRNPRD,RNCV,NVOR1,NVRN	PLOTS	13
	COMMON /TITL/ TITLE(6)	PLOTS	14
	COMMON /RJT/KARRAY2(7),AMYTRIX(3,3)	PLOTS	15
	INTEGER RNCV	PLOTS	16
C		PLOTS	17
C	THIS PROGRAM READS A SET OF 3 ANGLES AND PLOTS THE	PLOTS	18
C	CONFIGURATION AFTER ROTATING IT THROUGH THE INDICATED ANGLES.	PLOTS	19
C	FIRST ONE SIDE IS PLOTTED THEN IT IS REFLECTED THROUGH	PLOTS	20
C	ITS X-Z PLANE AND THE OTHER SIDE IS PLOTTED.	PLOTS	21
C	THE ROTATIONS ARE CARRIED OUT BY MATRIX MULTIPLICATION.	PLOTS	22
C	IT IS ALSO DETERMINED IF THE SUBPANELING AND CONTROL	PLOTS	23
C	POINTS ARE TO BE SHOWN.	PLOTS	24
C		PLOTS	25
C	LOGICAL SUBLINE,CPJNTZ	PLOTS	26
	DATA (KARRAY2(M),M=1,7)/20,3, 3,500,3,3,3/	PLOTS	27
	DATA PIE/3.14159/	PLOTS	28
	REWIND 3	PLOTS	29
9	REWIND 4	PLOTS	30
	REWIND 5	PLOTS	31

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# APPENDIX C. - Continued

32	PLOTS	
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C
      READ(3)NPAN,TITLE,TESTRTO
      HERE THE ANGLES OF ROTATION ARE READ; ROLL, PITCH, AND YAW,
      ARE READ.

      CALL ERASE
      WRITE(2,104)
      READ *,ROLL
      IF(ROLL.GT.360.)RETURN
      WRITE(2,105)
      READ *, PITCH
      WRITE(2,106)
      READ *,YAW
      WRITE(2,107)
      READ 112,SUBLINE
      WRITE(2,108)
      READ 112,CPOINTZ
      104 FORMAT(" INPUT THE ROLL ANGLE FOR THE AIRCRAFT",/
      2" (DEG), >360 TO STOP.")
      105 FORMAT("PITCH ANGLE")
      106 FORMAT("YAW ANGLE")
      107 FORMAT("SUBPANELING ? TRUE OR FALSE")
      108 FORMAT("CONTROL POINTS ? TRUE OR FALSE")
      112 FORMAT(L7)

C
      IF(TESTRTO.LT.1.00)GO TO 500
      IF(TESTRTO.GT.1.25)TESTRTO=1.25
      RYY=5.0/TESTRTO
      IXX2=INT(TESTRTO*3200)
      IXX1=INT((4096-IXX2)*.5)
      CALL SWINDO(IXX1,IXX2,1,3200)
      CALL DWINDO(-5.0,5.0,-RYY,RYY)
      GO TO 501

      500 CALL SWINDO(450,3200,1,3200)
      CALL DWINDO(-5.0,5.0,-5.0,5.0)
      501 CALL TURNIT(ROLL,PITCH,YAW)
      3 CALL MOVABS(100,3500)
      CALL ANMODE
  
```

# APPENDIX C. - Continued

```

WRITE(2,788)TITLE,ROLL,PITCH,YAW
788 FORMAT(8A10,/, " THE ROLL ANGLE IS ",F7.1,
1/, " THE PITCH ANGLE IS ",F7.1,
2/, " THE YAW ANGLE IS ",F7.1)
DO 300 NPANS=1,NPAN
DO 299 L=1,2
READ(3)NVR,NVCV,PDL,IQUANT,NVRNPRD,ISETS
ISET=ISETS
NVDRI=NVDRI+1
NVRN=NVRNPRD-NVDRI+1
IF(SUBLINE.A.PDL.LT.360)CALL VORSUB(5H LINES,ISETS,3,L)
IF(.NOT.SUBLINE.A.PDL.GT.360)CALL VORSUB(5H LINES,ISETS,3,L)
IF(SUBLINE.A.PDL.GT.360)CALL RYDLIN(L)
IF(.NOT.SUBLINE.A.PDL.LT.360)CALL VORSUB(5H EDGES,ISETS,3,L)
IF(PDL.GF.O.O.A.CPONTIZ)CALL CPOINTS(L)
IF(SUBLINE.A.PDL.GT.360)ISET=0
IF(L.EQ.2.OR.IQUANT.EQ.1)GO TO 300
CALL BACKUP(3,ISET+1)
299 CONTINUE
300 IF(L.EQ.2.A.ISET.EQ.0.OR.IQUANT.EQ.1.A.ISET.EQ.0)CALL SKPFILE(3,
1ISETS)
CALL MOVABS(100,3500)
CALL BIGER(II)
IF(II-1)9,9,3
END
C
SUBROUTINE VORSUB(WHICH,ISETS,IFILE,L)
COMMON /PRAMS/NVR,NVRNPRD,RNCV,NVDRI,NVRN
COMMON VORS(3,500)
C
C THIS SUBROUTINE PLOTS A FLAT MAJOR PANEL WITH
C CR WITHOUT SUBPANELING, OR A CURVED PANEL WITHOUT SUBPANELING.
C
CALL SAVE(IFUG,IFILE,IDUM,1,L,ISETS)
IF(WHICH.EQ.5H LINES)GO TO 1
N1=NVRN-1

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# APPENDIX C. - Continued

N2=NFOR	PLOTS	106
GO TO 2	PLOTS	107
1 N1=NFOR1	PLOTS	108
N2=1	PLOTS	109
2 CALL SAVE(IFUG,IFILE,NVRNPRD,2,L,ISETS)	PLOTS	110
DO 257 I=1,NVRN,N1	PLOTS	111
DO 257 J=1,NFOR1	PLOTS	112
I2=I+J-1+IFUG	PLOTS	113
IF(I2.LE.500)GO TO 3	PLOTS	114
10 I2=I2-500	PLOTS	115
CALL SAVE(IFUG,IFILE,NVRNPRD,2,L,ISETS)	PLOTS	116
IF(I2.GT.500)GO TO 10	PLOTS	117
3 IF(I2.EQ.1)CALL MOVEA(VORS(1,I2),VORS(3,I2))	PLOTS	118
257 CALL DRAWA(VORS(1,I2),VORS(3,I2))	PLOTS	119
DO 254 J=1,NFOR1,N2	PLOTS	120
IF(ISETS.EQ.1)GO TO 4	PLOTS	121
CALL BACKUP(IFILE,ISETS)	PLOTS	122
CALL SAVE(IFUG,IFILE,NVRNPRD,1,L,ISETS)	PLOTS	123
CALL SAVE(IFUG,IFILE,NVRNPRD,2,L,ISETS)	PLOTS	124
4 DO 254 I=1,NFOR1,NFOR1	PLOTS	125
I2=I+J-1+IFUG	PLOTS	126
IF(I2.LE.500)GO TO 5	PLOTS	127
CALL SAVE(IFUG,IFILE,NVRNPRD,2,L,ISETS)	PLOTS	128
I2=I2-500	PLOTS	129
5 IF(I2.EQ.1)CALL MOVEA(VORS(1,I2),VORS(3,I2))	PLOTS	130
258 CALL DRAWA(VORS(1,I2),VORS(3,I2))	PLOTS	131
RETURN	PLOTS	132
END	PLOTS	133
	PLOTS	134
SUBROUTINE REFLECT(N)	PLOTS	135
COMMON VORS(3,500)	PLOTS	136
THIS SUBROUTINE REFLECTS THE CONFIGURATION ACCROSS ITS X-Z PLANE.	PLOTS	137
DO 1 J=1,N	PLOTS	138
1 VORS(2,J)=-VORS(2,J)	PLOTS	139
RETURN	PLOTS	140
END	PLOTS	141

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PLOTS	142
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CALL ANMODE
WRITE(2,103)
READ 102, ENLARGE
IF(ENLARGE.EQ.3)YES=GO TO 1
I=1
RETURN

1 CALL VCURSR(ICCHAR,X1,Y1)
CALL VCURSR(ICCHAR,X2,Y2)
CALL FPA$F
RATIO=(X2-X1)/(Y2-Y1)
IF(RATIO.LT.1.25)GO TO 10
IX1=4F
IX2=4000
IY2=INT(4000/RATIO)
IY1=INT((3200-IY2)*.5)
GO TO 20

10 IY1=1
IY2=3200
IX2=INT(3200*RATIO)
IX1=INT((4098-IX2)*.5)

20 CALL SWINDO(IX1,IX2,IY1,IY2)
CALL DWINDO(X1,X2,Y1,Y2)
REWIND 3
REWIND 4
REWIND 5
READ(3)
I=2
RETURN

102 FORMAT(1A3)
103 FORMAT(//////," FOR ENLARGEM
END

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# APPENDIX C. - Continued

178 PLOTS  
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SUBROUTINE CPPOINTS(L)
  THIS SUBROUTINE PLOTS X'S ON THE CONTROL POINTS.
  COMMON VORS(3,500)
  COMMON/PRAMS/NVOR,NVRNPRD,RNCV,NVORI,NVRN
  INTEGER PRD,RNCV
  IF(L.EQ.2)CALL BACKUP(4,ISETS+1)
  READ(4)NVOR,RNCV,PRD,NNN,NVRNPRD,NN0,ISETS
  CALL SAVE(IFUG,4,IDUM,1,L,ISETS)
  CALL SAVE(IFUG,4,NNN,2,L,ISETS)
  DO 1 J=1,NN0,NVOR
    DO 1 I=1,NVOR
      K=(I+J-2)*4+IFUG
      K4=K+4
      CALL MOVEA(VORS(1,K+1),VORS(3,K+1))
      CALL DRAWA(VORS(1,K+2),VORS(3,K+2))
      CALL MOVEA(VORS(1,K+3),VORS(3,K+3))
      CALL DRAWA(VORS(1,K+4),VORS(3,K+4))
      1 IF(K4.EQ.500)CALL SAVE(IFUG,4,NNN,2,L,ISETS)
  RETURN
  END

```

200 PLOTS  
201 PLOTS  
202 PLOTS  
203 PLOTS  
204 PLOTS  
205 PLOTS  
206 PLOTS  
207 PLOTS  
208 PLOTS  
209 PLOTS  
210 PLOTS  
211 PLOTS  
212 PLOTS  
213 PLOTS

```

SUBROUTINE PNDLINE(L)
  THIS DEFINES A SET OF PARAMETERS SO A CURVED MAJOR
  PANEL WITH SUBPANELING CAN BE PLOTTED.
  INTEGER RNCV
  COMMON VORS(3,500)
  COMMON/PRAMS/NVOR,NVRNPRD,RNCV,NVORI,NVRN
  IF(L.EQ.2)CALL BACKUP(5,ISETS+1)
  READ(5)NVOR,RNCV,NVRNPRD,ISETS
  NVORI=NVOR+1
  NVRN=NVRNPRD-1
  CALL VORSUB(5,LINES,ISETS,5,L)
  RETURN
  END

```

# APPENDIX C. - Continued

```

C
C
SUBROUTINE SAVE(FUG,FILE,P,TEST,L,SETS)
  THIS SUBROUTINE MAKES IT POSSIBLE TO REDUCE THE CORE
  REQUIREMENTS OF THE PROGRAM.
  INTEGER FUG,FILE,P,TEST,SETS
  COMMON VORS(3,500)
  COMMON /PRAMS/NVOR,NVRNPRD,RNCV,NVORI,NVRN
  COMMON/ROT/KARRAY2(7),AMATRIX(3,3)
  IF(TEST.EQ.1)GO TO 1
  FUG=-500*TEST2
  IF(TEST2.EQ.SETS)RETURN
  TEST2=TEST2+1
  IF(TEST2.LT.SETS)N=500
  IF(TEST2.EQ.SETS)N=MOD(P,500)
  IF(N.EQ.0)N=500
  READ(FILE)((VORS(I,J),I=1,3),J=1,N)
  IF(L.EQ.2)CALL REFLECT(N)
  KARRAY2(4)=N
  CALL MATOPS(KARRAY2,AMATRIX,VORS,VORS)
  RETURN
  1 TEST2=FUG=0
  RETURN
  END
C

```

PLOTS 214  
 PLOTS 215  
 PLOTS 216  
 PLOTS 217  
 PLOTS 218  
 PLOTS 219  
 PLOTS 220  
 PLOTS 221  
 PLOTS 222  
 PLOTS 223  
 PLOTS 224  
 PLOTS 225  
 PLOTS 226  
 PLOTS 227  
 PLOTS 228  
 PLOTS 229  
 PLOTS 230  
 PLOTS 231  
 PLOTS 232  
 PLOTS 233  
 PLOTS 234  
 PLOTS 235  
 PLOTS 236

```

C
SUBROUTINE BACKUP(I,N)
  CAUSES FILE I TO BE BACKSPACED N RECORDS.
  DO 1 J=1,N
  1 BACKSPACE I
  RETURN
  END
C

```

PLOTS 237  
 PLOTS 238  
 PLOTS 239  
 PLOTS 240  
 PLOTS 241  
 PLOTS 242  
 PLOTS 243

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# APPENDIX C. - Continued

244	PLOTS		
245	PLOTS		
246	PLOTS		
247	PLOTS		
248	PLOTS		
249	PLOTS		
250	PLOTS		
251	PLOTS		
252	PLOTS		
253	PLOTS		
254	PLOTS		
255	PLOTS		
256	PLOTS		
257	PLOTS		
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271	PLOTS		
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273	PLOTS		
274	PLOTS		
275	PLOTS		
276	PLOTS		
277	PLOTS		
278	PLOTS		
279	PLOTS		

```

SUBROUTINE SKPFILE(I,N)
  CAUSES FILE I TO BE ADVANCED N RECORDS.
  DO 1 J=1,N
    1 READ(I)
  RETURN
END

SUBROUTINE TURNIT(RO,PO,YO)
  THIS SUBROUTINE DETERMINES THE MAYTRIX WHICH WILL ROTATE
  THE AIRCRAFT THROUGH THE ANGLES ROLL PITCH AND YAW AS
  THEY ARE NORMALLY DETERMINED.

  DIMENSION ZAX(3,3),XAX(3,3),R1(3,3),R2(3,3),VECTOR(3)
  DIMENSION KMULT(7),MULTV(7)
  COMMON /ROT/ XARRAY2(7),AMYTRIX(3,3)
  DATA(KMULT(M),M=1,7)/20,6*(3)/
  DATA(MULTV(M),M=1,7)/20,3,3,1,3,3,3/
  DATA PIE/3.14159/
  DATA XAX(1,2),XAX(1,3),XAX(2,1),XAX(3,1)/4*(0.0)/
  DATA ZAX(1,3),ZAX(2,3),ZAX(3,1),ZAX(3,2)/4*(0.0)/
  DATA R1(1,2),R1(1,3),R1(2,1),R1(3,1)/4*(0.0)/
  DATA R2(1,2),R2(2,1),R2(2,3),R2(3,2)/4*(0.0)/
  DATA XAX(1,1),ZAX(3,3),R1(1,1),R2(2,2)/4*(1.0)/
  1 AMYTRIX(1,1)=AMYTRIX(2,2)=AMYTRIX(3,3)=1.0
  AMYTRIX(1,2)=AMYTRIX(1,3)=AMYTRIX(2,1)=AMYTRIX(2,3)=0.0
  AMYTRIX(3,1)=AMYTRIX(3,2)=0.0
  VECTOR(1)=VECTOR(2)=0.0
  VECTOR(3)=1.0
  R=RO*PIE/180.
  P=-PO*PIE/180.
  Y=-YO*PIE/180.
  RTO=PIE*.5-R
  XAX(2,2)=XAX(3,3)=COS(R)
  SINE=SIN(R)
  XAX(2,3)=SINE
  XAX(3,2)=-SINE

```

# APPENDIX C. - Continued

```

280 ZAX(1,1)=ZAX(2,2)=COS(P)
281 SINE=SIN(P)
282 ZAX(1,2)=-SINE
283 ZAX(2,1)=SINE
284 R1(2,2)=R1(3,3)=COS(RTO)
285 SINE=SIN(RTO)
286 R1(2,3)=SINE
287 R1(3,2)=-SINE
288 CALL MATOPS(KMULT,XAX,AMYTRIX,AMYTRIX)
289 CALL MATOPS(KMULT,R1,AMYTRIX,AMYTRIX)
290 CALL MATOPS(KMULT,ZAX,AMYTRIX,AMYTRIX)
291 TRANSPOSE MAYTRIX R1.
292 R1(2,3)=-SINE
293 R1(3,2)=SINE
294 CALL MATOPS(KMULT,R1,AMYTRIX,AMYTRIX)
295 ZAX(1,1)=ZAX(2,2)=COS(Y)
296 SINE=SIN(Y)
297 ZAX(1,2)=-SINE
298 ZAX(2,1)=SINE
299 CALL MATOPS(MULTV,AMYTRIX,VECTOR,VECTOR)
300 A=VECTOR(1)
301 B=VECTOR(2)
302 C=VECTOR(3)
303 V=SQRT(B*B+C*C)
304 BOVRV=B/V
305 R1(2,2)=R1(3,3)=C/V
306 R1(2,3)=-BOVRV
307 R1(3,2)=BOVRV
308 R2(1,1)=R2(3,3)=V
309 R2(1,3)=-A
310 R2(3,1)=A
311 CALL MATOPS(KMULT,R1,AMYTRIX,AMYTRIX)
312 CALL MATOPS(KMULT,R2,AMYTRIX,AMYTRIX)
313 CALL MATOPS(KMULT,ZAX,AMYTRIX,AMYTRIX)
314 TRANSPOSE MAYTRIX R1 AND MAYTRIX R2.
315 R1(2,3)=BOVRV

```

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C



APPENDIX C. - Concluded

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PLOTS  
PLOTS  
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PLOTS  
PLOTS  
PLOTS

R1(3,2)=-BOVRV  
R2(1,3)=+A  
R2(3,1)=-A  
CALL MATOPS(KMULT,R2,AMATRIX,AMATRIX)  
CALL MATOPS(KMULT,R1,AMATRIX,AMATRIX)  
RETURN  
END

AST 100 - 6/8/75 PER COORD SHEET 164(REVISED), 712 LB/SEC ENGINES (AST-JP-2)

THE ROLL ANGLE IS 0.0  
THE PITCH ANGLE IS 0.0  
THE YAW ANGLE IS 0.0  
FOR ENLARGEMENT INPUT YES



(a) With panel camber displayed.



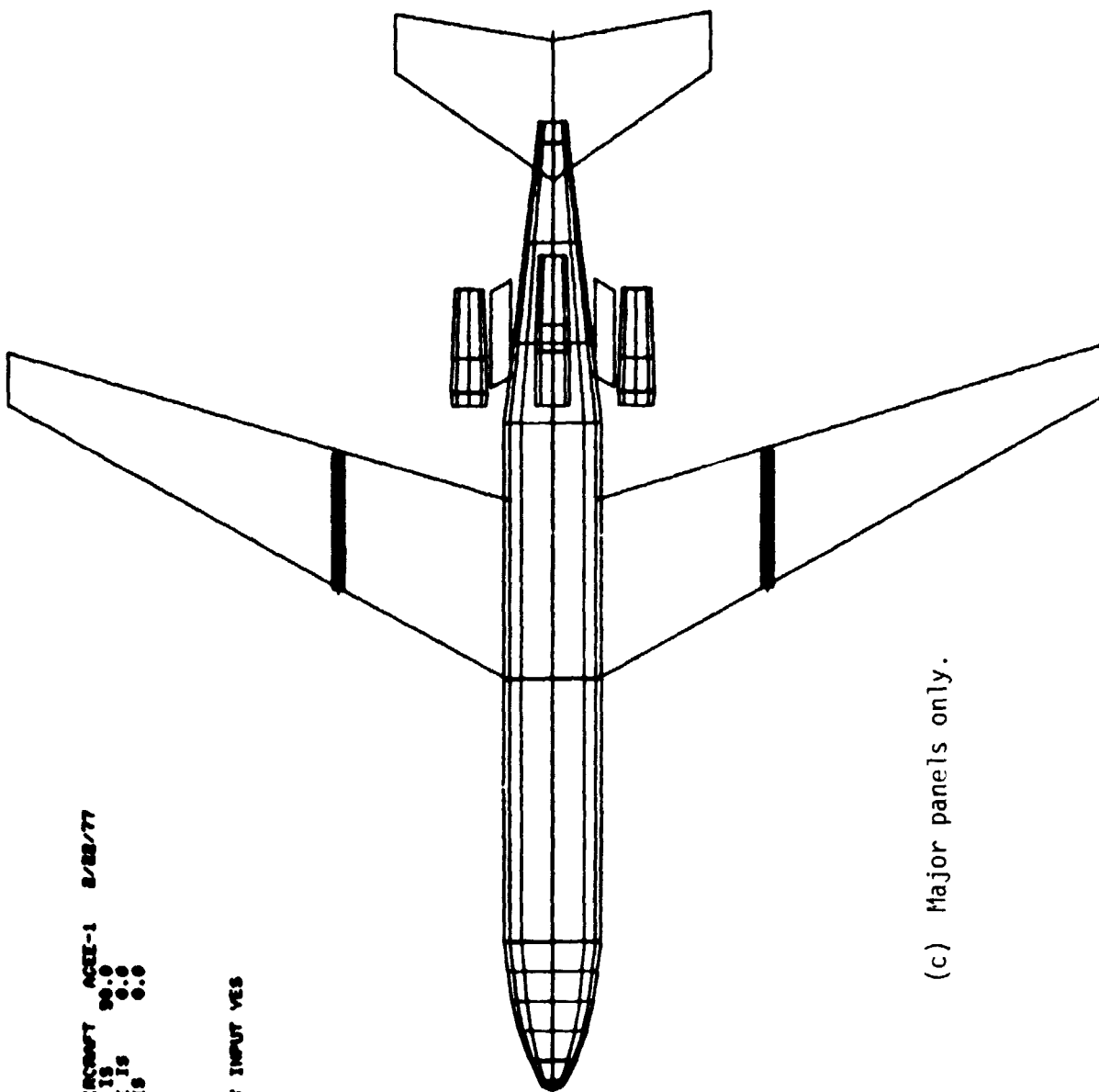
(b) Without panel camber displayed.

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Figure 1. - Display of aircraft configuration.

LAMINAR FLOW AIRCRAFT    ACEE-1    8/28/77  
 THE SOLL ANGLE IS    90.0  
 THE PITCH ANGLE IS    0.0  
 THE YAW ANGLE IS    0.0

FOR ENLARGEMENT INPUT YES

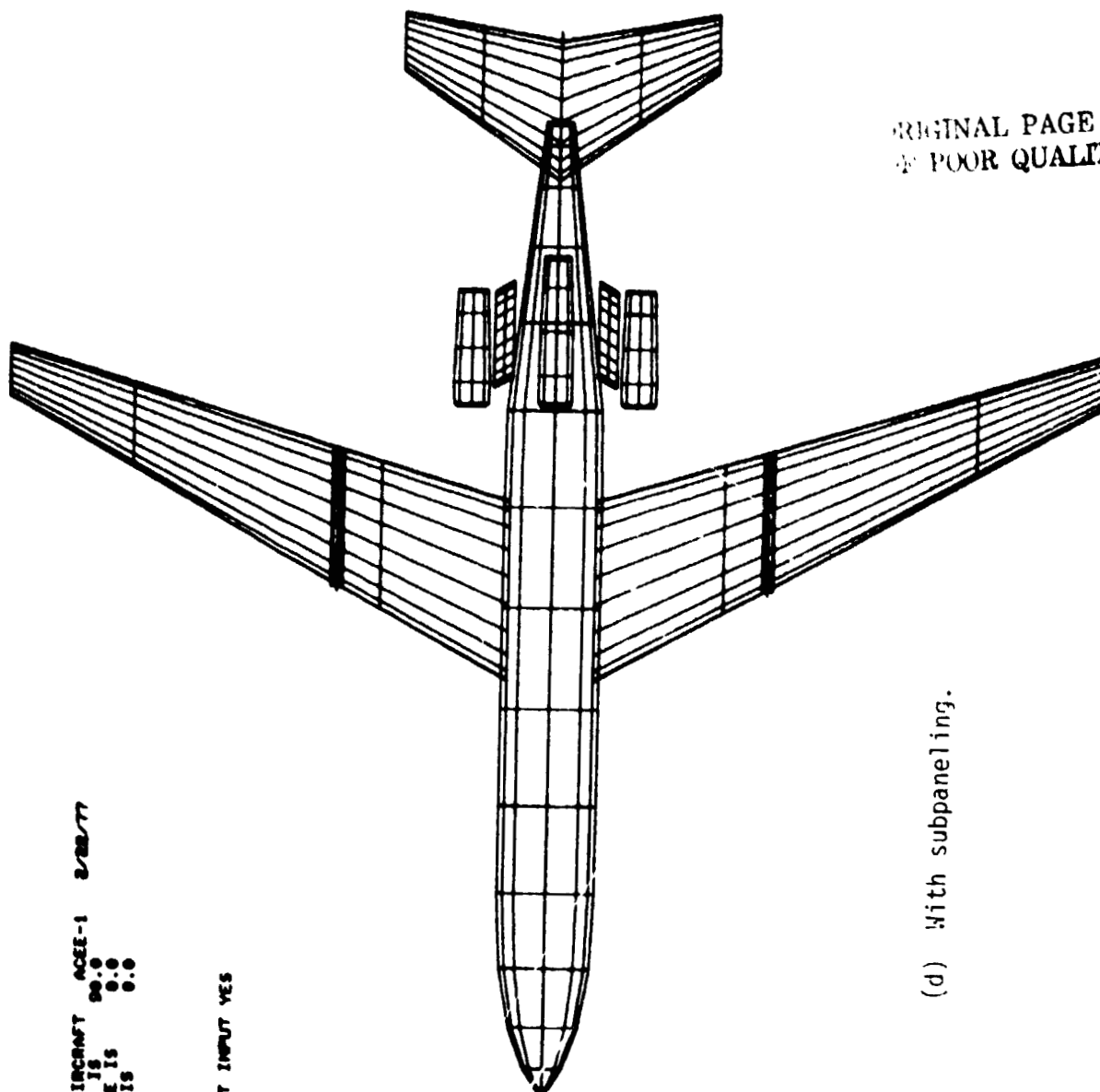


(c) Major panels only.

Figure 1. - Continued.

LARIMER FLOW AIRCRAFT    ACEE-1    2/22/77  
 THE ROLL ANGLE IS    0.0  
 THE PITCH ANGLE IS    0.0  
 THE YAW ANGLE IS    0.0

9  
 FOR ENLARGEMENT INPUT YES



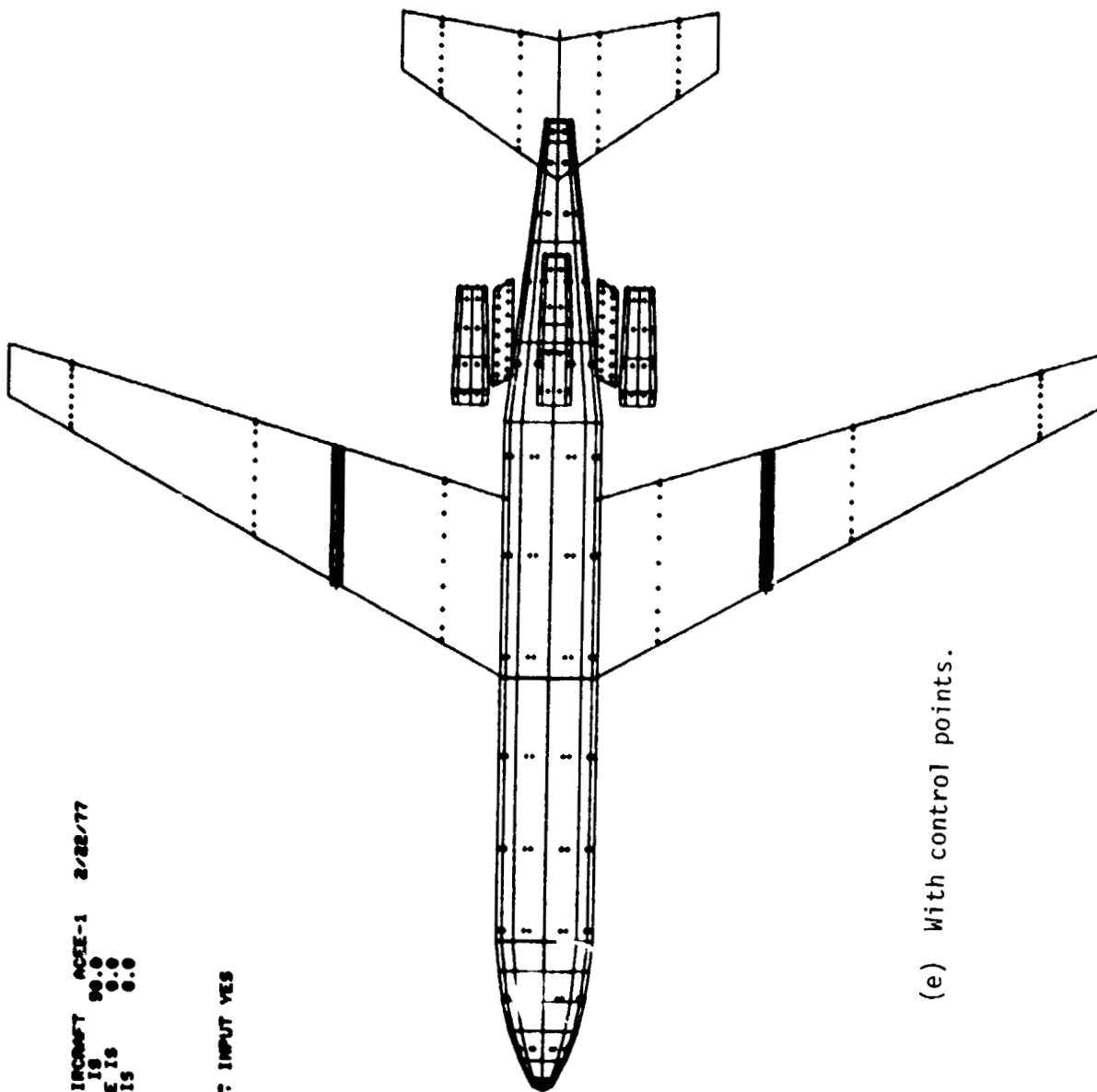
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(d) With subpaneling.

Figure 1. - Continued.

LAMINAR FLOW AIRCRAFT    ACIE-1    2/22/77  
 THE ROLL ANGLE IS    90.0  
 THE PITCH ANGLE IS    0.0  
 THE YAW ANGLE IS    0.0

FOR ENLARGEMENT: INPUT YES  
 ?

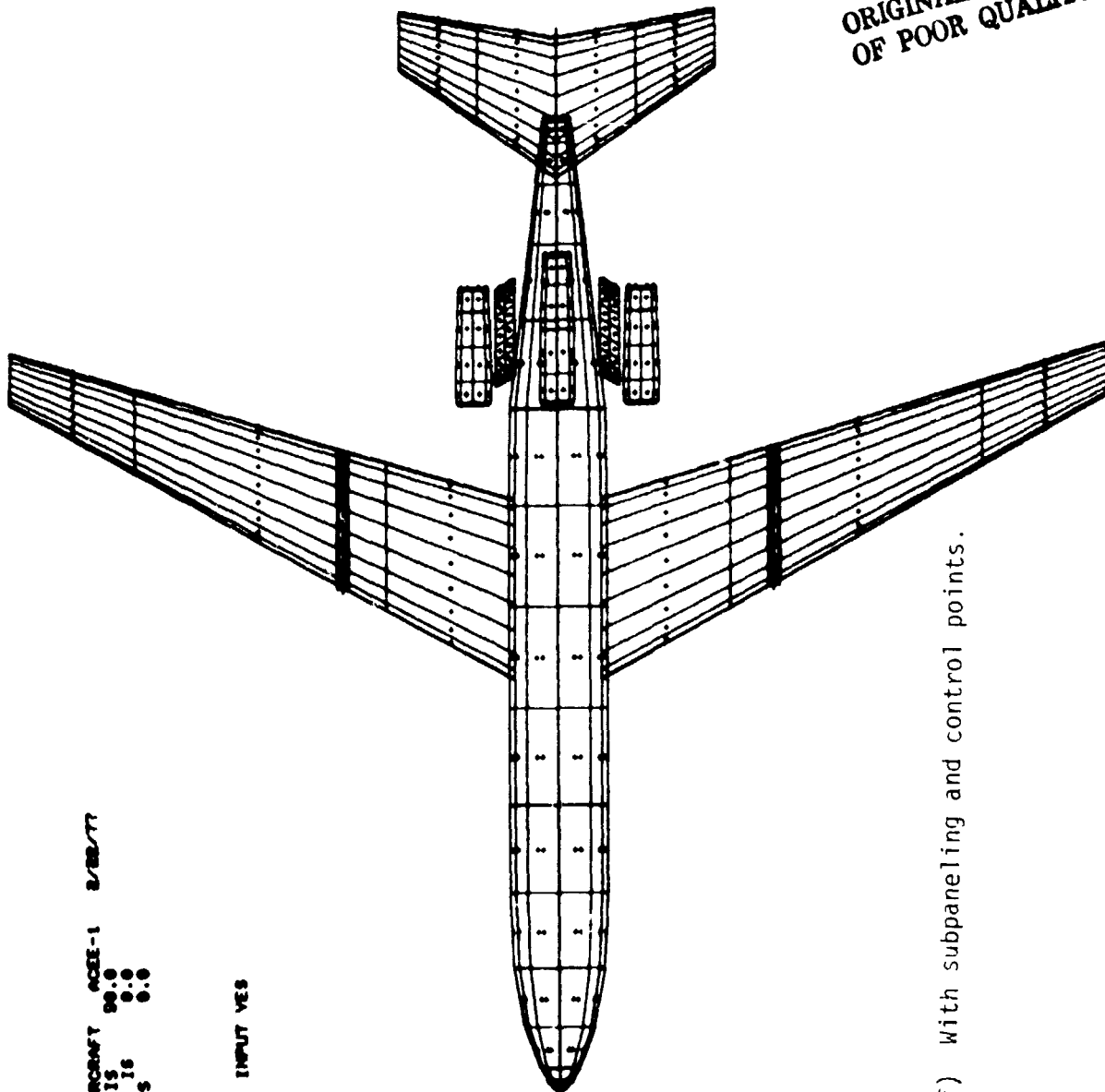


(e) With control points.

Figure 1. - Continued.

LANTIER FLOW AIRCRAFT ACEE-1 2/22/77  
 THE ROLL ANGLE IS 90.0  
 THE PITCH ANGLE IS 0.0  
 THE YAW ANGLE IS 0.0

FOR ENLARGEMENT INPUT YES



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(f) With subpaneling and control points.

Figure 1. - Concluded.

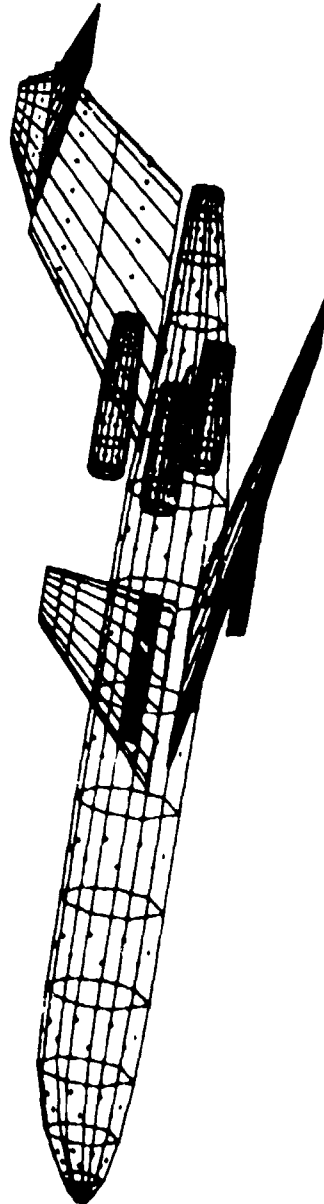
```
/GET,PLOTIT/UN=214737C
/CALL,PLOTIT(T=UORLAXX)
DO YOU WANT CAMBER IN THE PANELS ?
TRUE OR FALSE
? TRUE

    INPUT THE ROLL ANGLE FOR THE AIRCRAFT
    (DEG), >360 TO STOP.
? 11
PITCH ANGLE
? 12
YAW ANGLE
? 13
SUBPANELING ? TRUE OR FALSE
? TRUE
CONTROL POINTS ? TRUE OR FALSE
? TRUE
```

Figure 2. - Sample execution of program PLOTIT.

LAMINAR FLOW AIRCRAFT    ACEE-1    2/28/77  
THE ROLL ANGLE IS    11.0  
THE PITCH ANGLE IS    12.0  
THE YAW ANGLE IS    13.0

FOR ENLARGEMENT INPUT YES



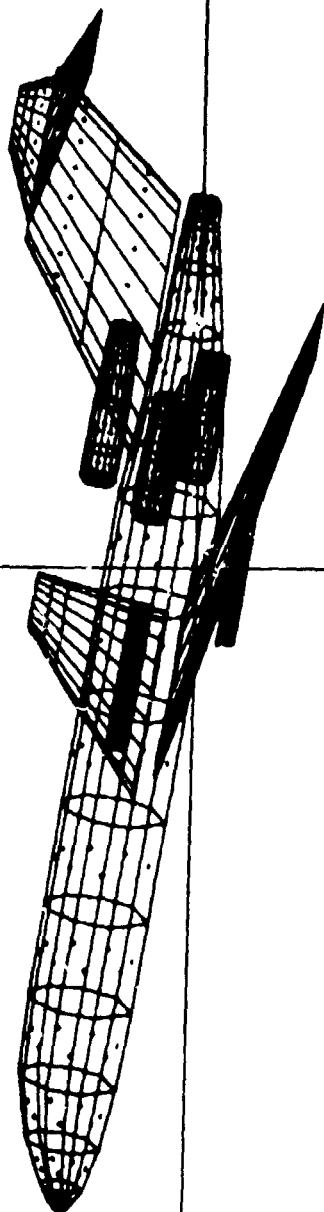
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Figure 3. - Plot from sample execution in figure 2.



LAUNCHER FLOW AIRCRAFT ACCE-1 8/28/77  
THE ROLL ANGLE IS 11.0  
THE PITCH ANGLE IS 18.0  
THE YAW ANGLE IS 13.0

FOR ENLARGEMENT INPUT YES

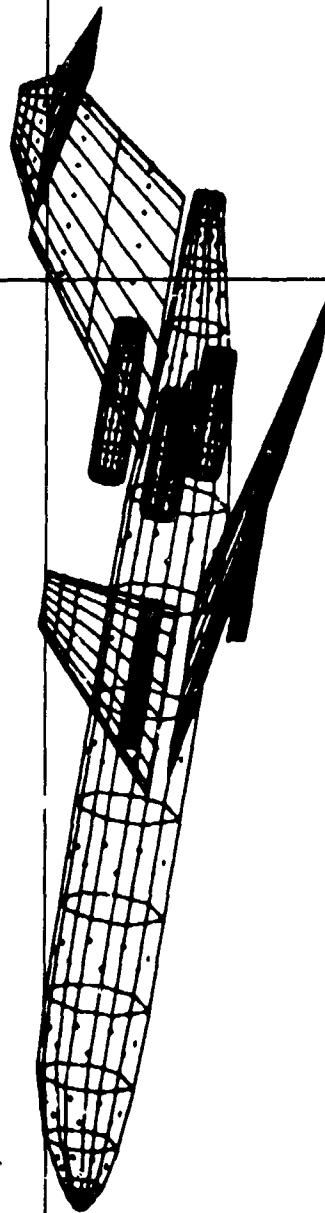


(a) First location of graphics cursor.

Figure 4. - Use of graphics cursor to enlarge a section of a configuration plot.

LANTIER FLOW AIRCRAFT    ACCE-1    9/28/77  
THE ROLL ANGLE IS    11.0  
THE PITCH ANGLE IS    12.0  
THE YAW ANGLE IS    13.0

FOR E-ARCMENT INPUT YES



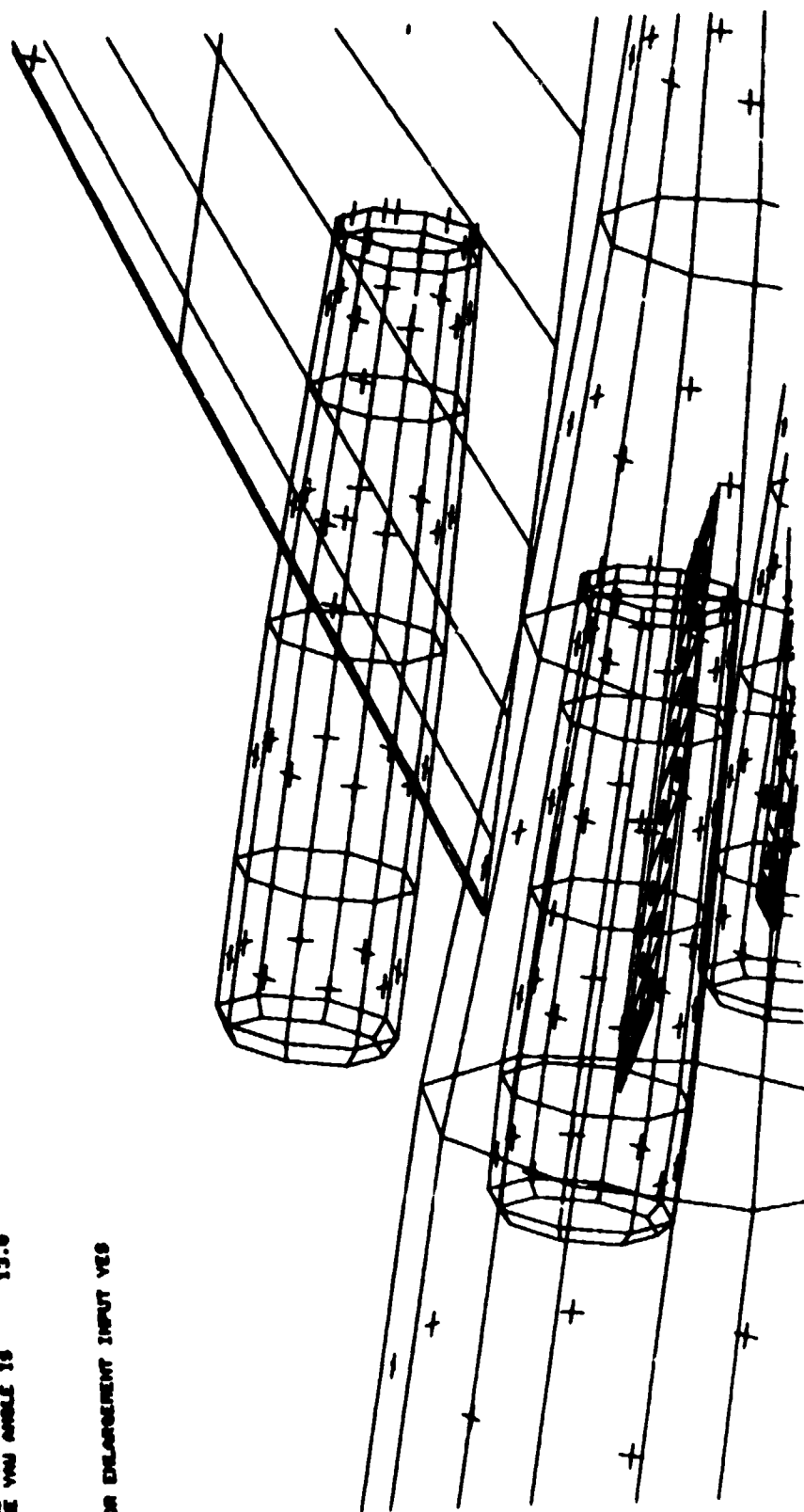
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(b) Second location of graphics cursor.

Figure 4. - Continued.

LASTING FLOW AIRCRAFT ACIE-1 8/25/77  
 THE ROLL ANGLE IS 11.0  
 THE PITCH ANGLE IS 12.0  
 THE YAW ANGLE IS 13.0

FOR ENLARGEMENT INPUT YES



(c) Enlarged plot resulting from cursor locations in figure 4 part (a) and part (b).

Figure 4. - Concluded.